

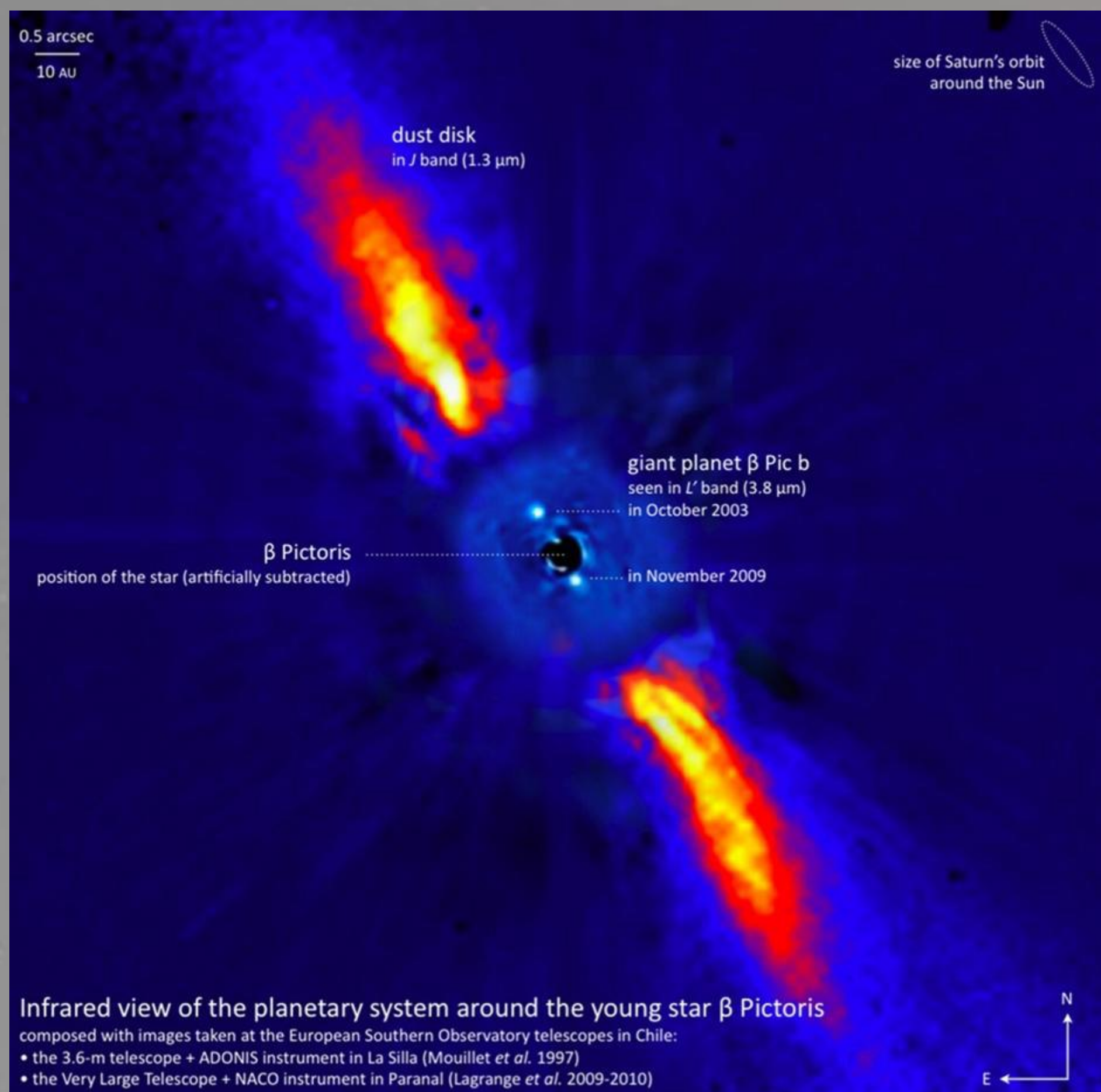
Beta Pictoris transit with PicSat

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The objective of the PicSat space mission is to study the β Pic b transit forecasted in summer 2017. The idea of this demonstrator is to spatialize a 3U CubeSat using a 35mm effective aperture and a single pixel avalanche photodiode. To guarantee photometric precision and CubeSat stabilization, the Attitude Determination and Control System is loop controlled with a double-axis piezo-actuator. The piezo follows the maximum intensity of the signal in the focal plane to compensate the jitter of the pointing system. PicSat uses a single-mode fiber to guide the stellar light from the focal plane to the photodiode.



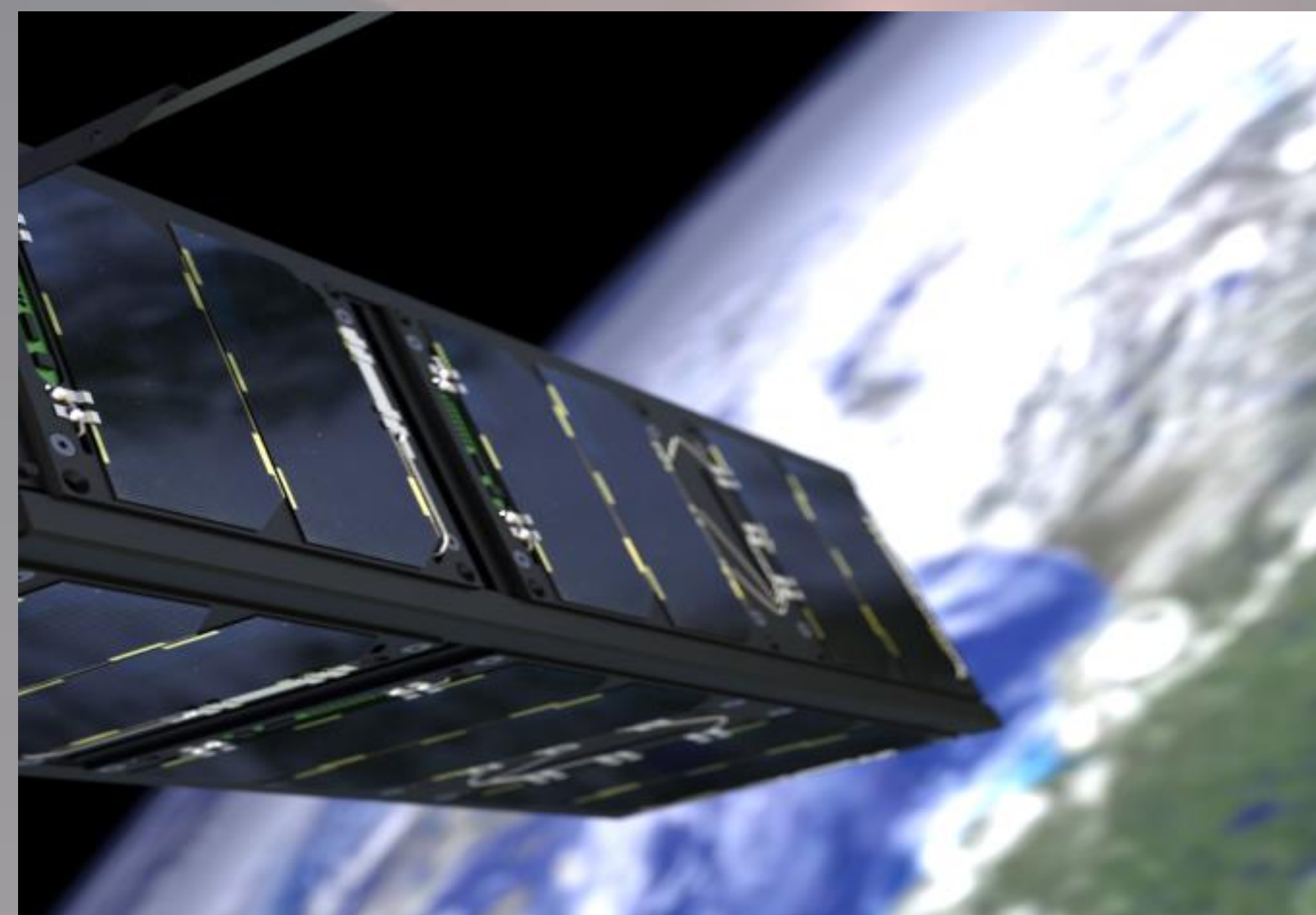
Beta Pictoris

β Pic is a 20 Myr old and 63 light-years away A5 star.

A large circumstellar disk is seen edge-on, in which an exoplanet, β Pic b was discovered in 2008 with VLT-NACO adaptive optics.

This planet -a Jupiter like gas giant- estimated mass is about 8-10 M_J and has a semi-major axis of 8 au.

CubeSat Standard

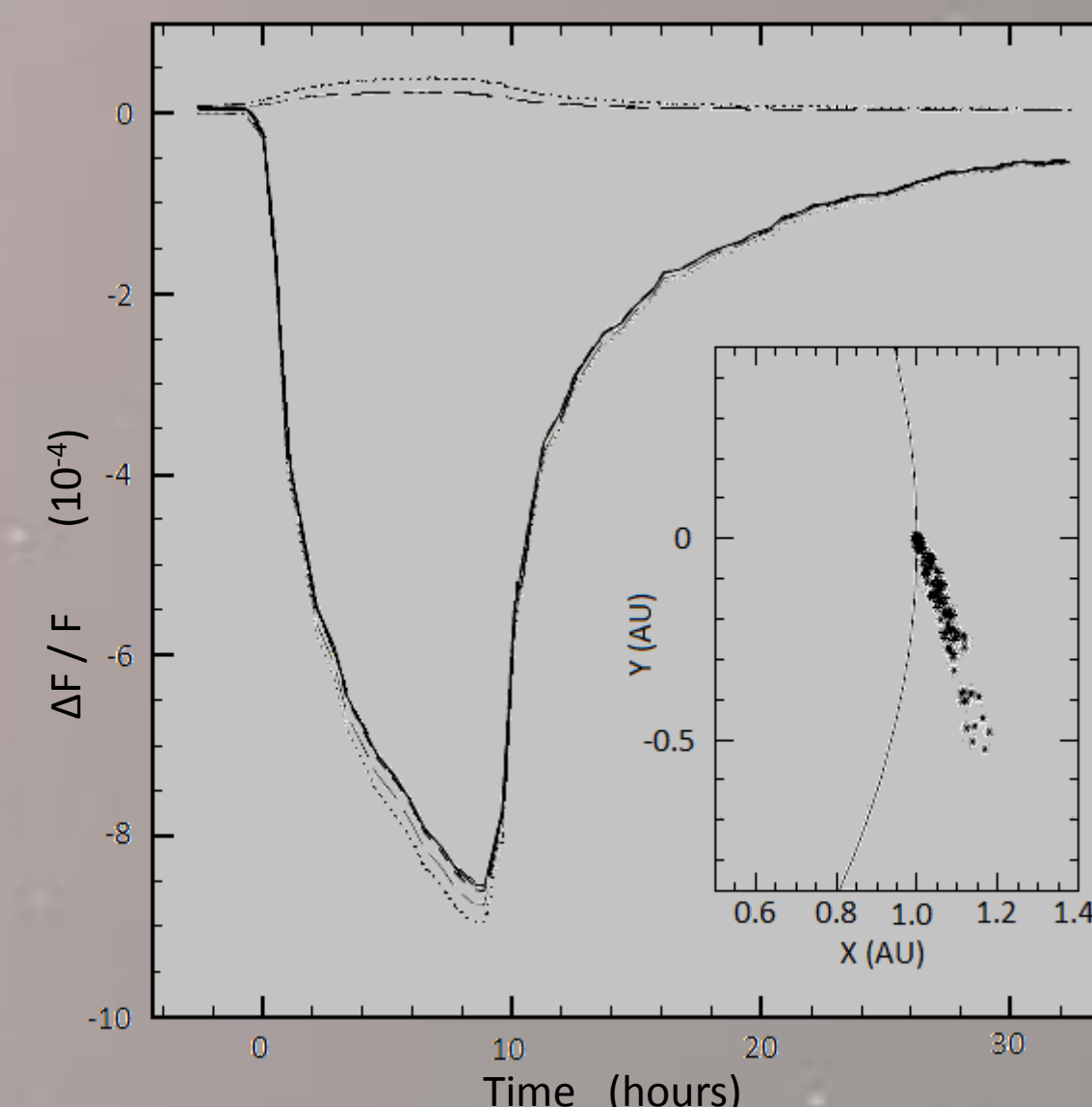


Developped in California and Stanford univeristies in 1999, CubeSat's objectives is to provide a standard for the design of small satellites, in order to reduce cost, development time and increase accessibility to space. For now, 340 CubeSat were launched.

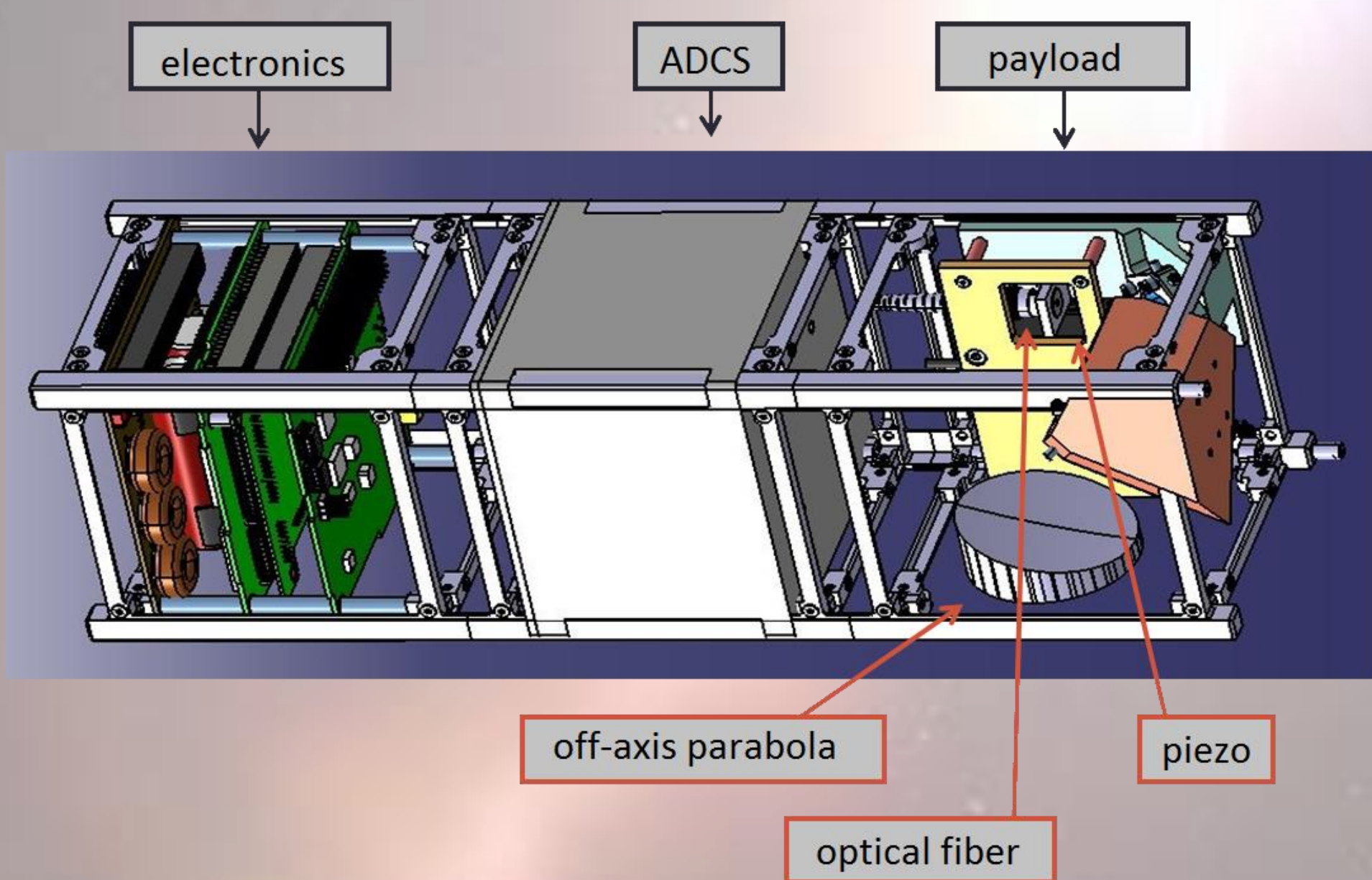
Expected Transits

Orbital parameters of β Pic b considering the 1981 photometric event as a transit coupled with 2009-2013 observations, with a low eccentricity orbit ($e \approx 0.1$, periode ≈ 18 yr) Next transit maximum probability is mid 2017.

In case of a high eccentricity ($e \approx 0.3$, periode ≈ 36 yr) the next transit will be in winter-spring 2018.



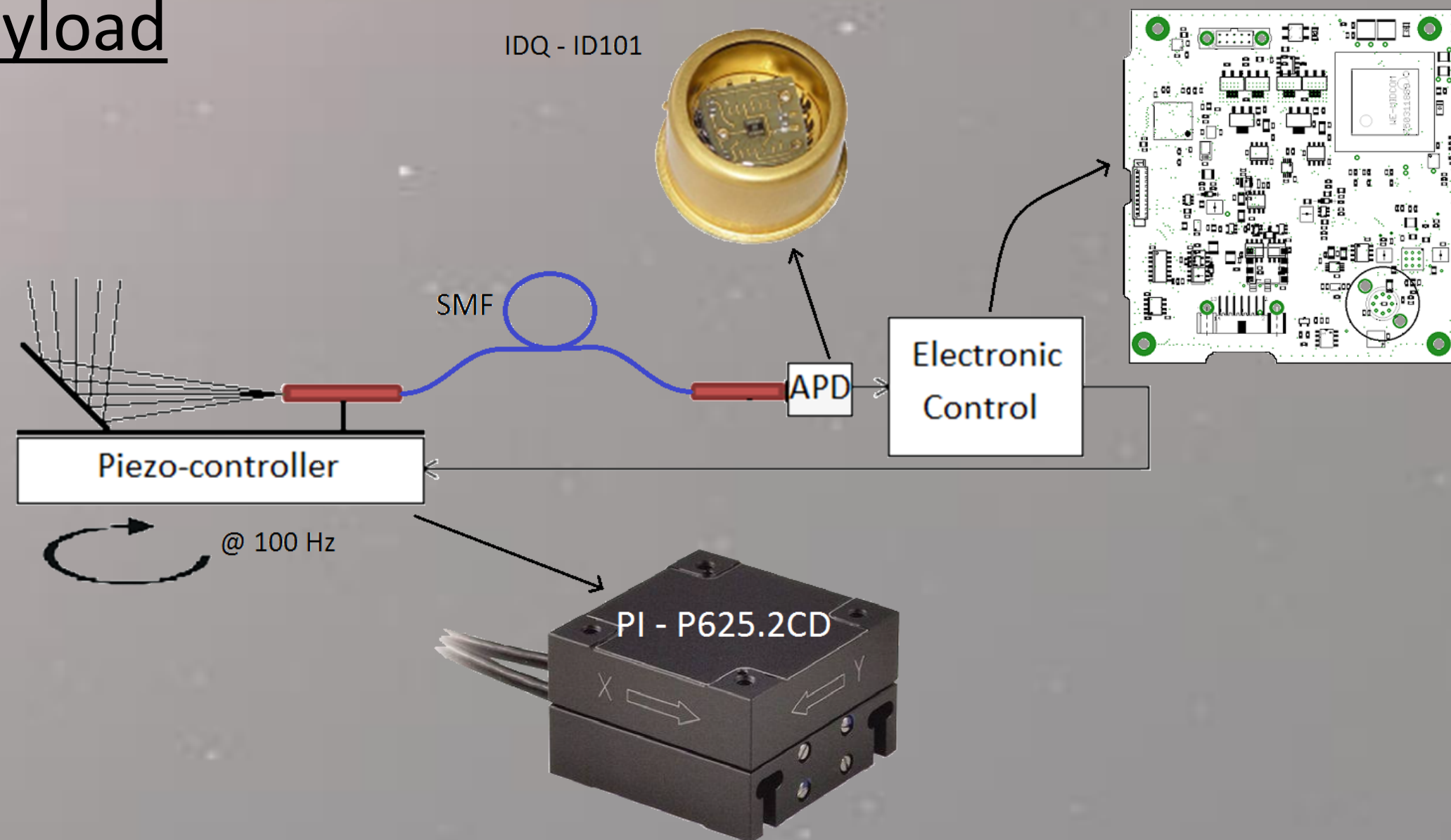
Simulation of photometric variations during cometary occultation. The insert is a view from the top when the comet is crossing the line of sight ($Y = 0$) at the periastron. Light curve presents the very specific "rounded triangular" shape. Production rate at 1 AU is 2.10^6 kg.s^{-1}



The satellite is based on a 3 units CubeSat platform (10x10x34cm) with 1U for electronics, 1U for ADCS and 1U for our payload. A prototype of the payload is currently being tested in Meudon, electronics components and ADCS are off-the-shelf systems.

The telescope is a 50mm F/2.7 parabola (and 2 reflecting mirrors) coupled with a single mode fiber carrying the light to an Avalanche PhotoDiode.

Payload



The payload will track the star by moving the fiber head in the focal plane. The Single Mode Fiber (SMF) is mounted on a 2 axes piezo stage that modulates the position around the star with a 100Hz frequency.

The photometric error budget gives a total error of 168 ppm/hour considering a 35mm diaphragm, a total transmission of 20%, a quantum efficiency of 30% for a Vmagnitude = 3.86 star.

	Assumption	Error	Error (ppm/hour)
Photon noise	8.10^4 e/s	$1/\sqrt{N_{\text{photon}}}$	60
Readout noise	0	0	0
Dark current	10^3 e/s	$\sqrt{N_{\text{current}}/N_{\text{photon}}}$	0.01
Scattered light	150 e/s	$N_{\text{moon}}/N_{\text{photon}}$	77
Thermal stability	0.01°C	0.4% per °C	40
Voltage stability	100µV	20% per volt	20
Pointing stability	5% @ 100Hz	$\Delta \ln j \times \sqrt{t} \times 100 \text{ Hz}$	83
Total error			168

